

IN THE CLAIMS

Please amend the claims as follows:

Claim 1 (Original): A rotary turbojet blade that is to be subjected to a longitudinal gas flow, said blade comprising a plurality of blade sections extending along a line of the centers of gravity of said blade sections between a base and a tip of said blade, said blade being defined longitudinally between a leading edge and a trailing edge, said blade presenting along a radial axis of said turbojet a bottom portion, an intermediate portion, and a top portion, said bottom portion extending radially from said blade base to a bottom limit of said intermediate portion, and said top portion extending radially from a top limit of said intermediate portion to said blade tip, wherein said bottom portion presents a longitudinal angle of inclination for a leading edge line, said intermediate portion presents a backward longitudinal angle of inclination for said leading edge line, and said top portion presents a backward longitudinal angle of inclination for said leading edge line and a tangential angle of inclination for said line of the centers of gravity of the blade sections in a direction opposite to the direction of rotation of said blade.

Claim 2 (Currently Amended): A blade according to claim 1, wherein said bottom limit of the intermediate portion of the blade lies in the range 40% to 75% of the radial height of said blade between its base and its tip, and said bottom limit of the intermediate portion of the blade is a point of the leading edge having a minimum abscissa.

Claim 3 (Original): A blade according to claim 1, wherein the longitudinal angle of inclination of the leading edge line of said bottom portion lies in the range -5° to 15° relative to said radial axis of the turbojet.

Claim 4 (Original): A blade according to claim 1, wherein the backward longitudinal angle of inclination of the leading edge line of the intermediate portion lies in the range 5° to 20° relative to said radial axis of the turbojet.

Claim 5 (Original): A blade according to claim 1, wherein the backward longitudinal angle of inclination of the leading edge line of said top portion lies in the range 20° to 50° , and the tangential angle of inclination of the line of the centers of gravity of the blade sections of said top portion lies in the range 20° to 50° relative to said radial axis of the turbojet.

Claim 6 (Original): A blade according to claim 1, wherein said bottom portion further presents a tangential angle of inclination for the line of the centers of gravity of the blade sections lying in the range -5° to 15° relative to said radial axis of the turbojet.

Claim 7 (Original): A blade according to claim 1, wherein said intermediate portion further presents a tangential angle of inclination for the line of the centers of gravity of the blade sections lying in the range -5° to 15° relative to said radial axis of the turbojet.

Claim 8 (Original): A blade according to claim 1, wherein said top portion further comprises a top zone extending radially to said blade tip in which the leading edge line presents a forward longitudinal angle of inclination.

Claim 9 (Original): A rotary turbojet machine for passing a flow of gas, including a plurality of blades according to claim 1.

Claim 10 (Original): A machine according to claim 9, constituting a turbojet fan.

Claim 11 (Original): A machine according to claim 9, constituting a turbojet compressor.

Claim 12 (New): A rotary turbojet blade that is to be subjected to a longitudinal gas flow, said blade comprising a plurality of blade sections extending along a line of the centers of gravity of said blade sections between a base and a tip of said blade, said blade being defined longitudinally between a leading edge and a trailing edge, said blade presenting along a radial axis of said turbojet a bottom portion, an intermediate portion, and a top portion, said bottom portion extending radially from said blade base to a bottom limit of said intermediate portion, and said top portion extending radially from a top limit of said intermediate portion to said blade tip, wherein said bottom portion presents a longitudinal angle of inclination for a leading edge line, said intermediate portion presents a backward longitudinal angle of inclination for said leading edge line, said top portion presents a backward longitudinal angle of inclination for said leading edge line and a tangential angle of inclination for said line of the centers of gravity of the blade sections in a direction opposite to the direction of rotation of said blade, and said top portion further comprises a top zone extending radially to said blade tip in which the leading edge line presents a forward longitudinal angle of inclination.

Claim 13 (New): A blade according to claim 12, wherein said bottom limit of the intermediate portion of the blade lies in the range 40% to 75% of the radial height of said blade between its base and its tip.

Claim 14 (New): A blade according to claim 12, wherein the longitudinal angle of inclination of the leading edge line of said bottom portion lies in the range -5° to 15° relative to said radial axis of the turbojet.

Claim 15 (New): A blade according to claim 12, wherein the backward longitudinal angle of inclination of the leading edge line of the intermediate portion lies in the range 5° to 20° relative to said radial axis of the turbojet.

Claim 16 (New): A blade according to claim 12, wherein the backward longitudinal angle of inclination of the leading edge line of said top portion lies in the range 20° to 50° , and the tangential angle of inclination of the line of the centers of gravity of the blade sections of said top portion lies in the range 20° to 50° relative to said radial axis of the turbojet.

Claim 17 (New): A blade according to claim 12, wherein said bottom portion further presents a tangential angle of inclination for the line of the centers of gravity of the blade sections lying in the range -5° to 15° relative to said radial axis of the turbojet.

Claim 18 (New): A blade according to claim 12, wherein said intermediate portion further presents a tangential angle of inclination for the line of the centers of gravity of the blade sections lying in the range -5° to 15° relative to said radial axis of the turbojet.